

aircraft applications where the location of the thrust augmenting ejectors on the wings necessitates compact geometries. The experiments described in this Note were not designed to study the acoustic interactions which were, in fact, observed by chance. The previous comments are therefore speculative in nature and have been offered in the interest of stimulating research.

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Errata

A Finite-Element Method for Calculating Aerodynamic Coefficients of a Subsonic Airplane

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THE sentence preceding Eq. (12) should read: For an axisymmetric body with angle of attack α and yaw angle β , the pressure coefficient at surface (r, θ, x) by neglecting high-order term $(u_x/V_\infty)^2$, is

$$C_p = - \frac{2u_x \cos \alpha \cos \beta}{V_\infty} + \frac{2 \sin \alpha \cos \beta}{V_\infty \cos \theta_0} \times [u_r \cos(\theta - \theta_0) - u_\theta \sin(\theta - \theta_0)] - \frac{u_r^2 + u_\theta^2}{V_\infty^2} \quad (12)$$

Both the denominators of Eqs. (A4) and (A9) should read 2π instead of 4π .

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